

Art Unit: 3641

With regard to item 2 of the printer rush, please see the included lines that clarify issues directed to the portions of the specification covered by stamp markings.

The last line of page 1 should read as follows:

experienced in missile systems. These high axial loads during launch of an artillery shell

The first line of page 2 should read as follows:

would cause movement and deformation of the propellant in a rocket motor which could

The last two lines of page 2 should read as follows:

ignition of the rocket motor for a predetermined time after launch of the shell wherein the rocket motor comprises a combustion chamber, a plenum chamber located at one end of

The last two lines of page 3 should read as follows:

Rocket motor ignition may be effected by a pyrotechnic delay through a rocket motor nozzle plug of the type commonly employed within rocket assisted shells to seal off

The first line of page 4 should read as follows:

the combustion chamber from the high temperature and high pressure gases present in the

The first line of page 5 should read as follows:

allowing a relative reduction in the thickness of the plate. This has the advantage that more

The last line of page 6 should read as follows:

layers, each comprising a PTFE substrate coated with between 4 to 6 microns of vapour

The last line of page 7 should read as follows:

when the base bleed propellant 16 has substantially finished burning. The transfer charge

The first line of page 8 should read as follows:

25 is designed to burn to release sufficiently hot combustion products into the plenum

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Claim 4 should read as follows:

4. A long range artillery shell as claimed in Claim 3 wherein the load transfer means consists of a load ring arranged concentric with and abutting the perforated plate.